

substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

■ Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. Section 39.13 is amended by adding a new airworthiness directive to read as follows:

2003-09-02 Pratt & Whitney: Amendment 39-13131. Docket No. 2002-NE-15-AD.

Applicability

This airworthiness directive (AD) is applicable to Pratt and Whitney (PW) models PW4164, PW4168, and PW4168A turbofan engines, serial numbers P733301 through P733500, that have HPC inner case assembly part number (P/N) 53H272-01 installed. These engines are installed on, but not limited to Airbus Industrie A330 airplanes.

Note 1: This AD applies to each engine identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For engines that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an

alternative method of compliance in accordance with paragraph (g) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance

Compliance with this AD is required as indicated, unless already done.

To prevent engine power loss during takeoff due to high pressure compressor (HPC) surge, do the following:

(a) Borescope-inspect in accordance with paragraphs 1.A. through 1.I of the Accomplishment Instructions, Borescope Inspection of Pratt & Whitney (PW) Alert Service Bulletin (ASB) PW4G-100-A72-170, Revision 3, dated January 31, 2003, using schedules in the following Table 1:

TABLE 1.—INSTALLED-ENGINE INITIAL BORESCOPE INSPECTION SCHEDULE

Accumulated engine cycles on the effective date of this AD:	Initial borescope-inspect 14th and 15th stage rubstrips:
(1) Fewer than 900 cycles-since-new (CSN) or cycles-since-refurbishment (CSR) of the HPC inner rear case assembly.	Before accumulating 1,500 CSN or CSR, whichever occurs later.
(2) 900 or more CSN or CSR	Within 600 cycles-in-service (CIS) after the effective date of this AD.

(b) Perform the applicable action as specified in the following Table 2:

TABLE 2.—BORESCOPE INSPECTION RESULTS AND REQUIRED ACTION

Results observed at last borescope inspection:	Action:
(1) 14th stage rubstrip shows wear through to the parent material of the stator, with the circumferential length of the wear being 1.0" or more.	Remove engine from service within 250 CIS.
(2) 15th stage rubstrip shows wear through to the parent material of the stator, with the circumferential length of the wear being 2.0" or more.	Remove engine from service within 250 CIS.
(3) 14th stage rubstrip shows wear through to the parent material of the stator, with the circumferential length of the wear being less than 1.0".	Reinspect every 300 CIS.
(4) 15th stage rubstrip shows wear through to the parent material of the stator, with the circumferential length of the wear being less than 2.0".	Reinspect every 300 CIS.
(5) Both 14th and 15th stage rubstrips show no wear through to the parent material of the stator.	Reinspect every 600 CIS.

Airplanes With Two Affected Engines Installed

(c) For airplanes with two affected engines installed, perform the actions as specified in following Table 3:

TABLE 3.—BORESCOPE INSPECTION RESULTS AND REQUIRED ACTIONS FOR TWO AFFECTED ENGINES

First engine borescope results:	Second engine borescope schedule:	Second engine borescope results:	Action:
(1) Wear as specified in Table 2, Steps (1) through (4).	Within 10 CIS since borescope inspection of first engine.	Wear as specified in Table 2, Steps (1) through (4).	Remove either the first or second engine from service within 25 CIS since borescope inspection of the second engine and replace with a serviceable engine. Remove or re-inspect in accordance with Table 2 for remaining engine and if still applicable perform actions as specified in paragraph (c).