

planning time, or time necessitated by other administrative actions.

Regulatory Impact

The regulations proposed herein would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this proposal would not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this proposed regulation (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by removing amendment 39-9952 (62 FR 9359, March 3, 1997), and by adding a new airworthiness directive (AD), to read as follows:

Boeing: Docket 2002-NM-66-AD.

Supersedes AD 97-05-08, Amendment 39-9952.

Applicability: All Model 727 series airplanes, certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the

requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (s)(1) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent fatigue cracking of the forward support fitting of the number 1 and number 3 engines, which could result in failure of the support fitting and consequent separation of the engine from the airplane, accomplish the following:

Restatement of Requirements of AD 97-05-08

Inspections

(a) Within 100 days or 600 flight cycles after March 18, 1997 (the effective date of AD 97-05-08, amendment 39-9952), whichever occurs first, accomplish paragraphs (a)(1), (a)(2), and (a)(3) of this AD, in accordance with Boeing Service Bulletin 727-54A0010, Revision 4, dated January 30, 1997.

(1) Perform a visual inspection to detect cracks of the upper and lower flanges, and the vertical web of the forward support fitting of the number 1 and number 3 engines, in accordance with Part 1—Pre-Modification Inspections of the Accomplishment Instructions of the service bulletin.

(2) Perform a high frequency eddy current (HFEC) inspection to detect cracks of the forward flange of the support fitting adjacent to the collars of two fasteners of the number 1 and number 3 engines, in accordance with Part 1—Pre-Modification Inspections of the Accomplishment Instructions of the service bulletin.

(3) Perform a detailed inspection to detect cracks of the upper and lower flanges adjacent to six fasteners of the fitting of the number 1 and number 3 engines, in accordance with Part 1—Pre-Modification Inspections of the Accomplishment Instructions of the service bulletin.

Note 2: For the purposes of this AD, a detailed inspection is defined as: "An intensive visual examination of a specific structural area, system, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at intensity deemed appropriate by the inspector. Inspection aids such as mirror, magnifying lenses, etc., may be used. Surface cleaning and elaborate access procedures may be required."

(b) If no crack is detected during the inspections required by paragraph (a) of this AD, repeat those inspections thereafter at intervals not to exceed 100 days or 600 flight cycles, whichever occurs first, until the initial inspections required by paragraph (d) of this AD have been accomplished.

(c) If any crack is detected during any inspection required by paragraph (a) of this AD, prior to further flight, repair the forward support fitting in accordance with a method approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA.

New Requirements of This AD

Note 3: Where there are differences between the service bulletin and this AD, this AD prevails.

Inspections: All Airplanes

(d) For all airplanes: Within 600 flight cycles or 100 days after the effective date of this AD, whichever occurs first, inspect the forward support fitting of the number 1 and number 3 engines, as specified in paragraphs (d)(1), (d)(2), (d)(3), (d)(4), and (d)(5) of this AD, in accordance with Part I of the Accomplishment Instructions of Boeing Service Bulletin 727-54A0010, Revision 6, dated August 23, 2001. Accomplishment of these initial inspections terminates the inspection requirements of paragraphs (a) and (b) of this AD.

(1) Perform a general visual inspection to detect corrosion and cracking of the fittings in areas inboard of the side of the body, in accordance with Figure 1 of the service bulletin. If any corrosion is found, before further flight, remove the corrosion in accordance with Figure 5 of the service bulletin, and then perform a general visual inspection to detect cracking of the area, in accordance with the service bulletin.

Note 4: For the purposes of this AD, a general visual inspection is defined as: "A visual examination of an interior or exterior area, installation, or assembly to detect obvious damage, failure, or irregularity. This level of inspection is made from within touching distance unless otherwise specified. A mirror may be necessary to enhance visual access to all exposed surfaces in the inspection area. This level of inspection is made under normally available lighting conditions such as daylight, hangar lighting, flashlight, or droplight and may require removal or opening of access panels or doors. Stands, ladders, or platforms may be required to gain proximity to the area being checked."

(2) Perform a HFEC inspection to detect cracking of the upper and lower horizontal flanges and post tangs of the fittings from inside the airplane, in accordance with Figure 1 of the service bulletin.

(3) Perform a general visual inspection to detect cracking and corrosion of the fittings in areas outboard of the side of the body, in accordance with Figure 1 of the service bulletin. If any corrosion is found, before further flight, remove the corrosion in accordance with Figure 5 of the service bulletin, and perform a general visual inspection to detect cracking of the area, in accordance with the service bulletin.

(4) Perform a detailed inspection to detect cracking and corrosion of the web in areas outboard of the side of the body, in accordance with Figure 1 of the service bulletin. If any corrosion is found, before further flight, remove the corrosion in accordance with Figure 5 of the service bulletin, and perform thickness measurements and detailed and HFEC inspections of the vertical web inboard and outboard of the side of the body to detect corrosion and cracking, in accordance with Figure 2 of the service bulletin.

(5) Perform detailed and HFEC inspections to detect cracking of the upper and lower