

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by removing Amendment 39–12793 (67 FR

43227, June 27, 2002), and by adding a new airworthiness directive to read as follows:

2002–13–05 R1 MD Helicopters, Inc.: Amendment 39–13022, Docket No. 2001–SW–40–AD. Revises AD 2002–13–05, Amendment 39–12793.

Applicability: The following MD Helicopters, Inc. helicopter models, certificated in any category:

Helicopter model	With	Modified In accordance with
(1) 369D, 369E, 369F, 369FF	Tailboom, serial number (S/N) 5001–5032	Aerometals Supplemental Type Certificate (STC) SH5055NM
(2) 369D and 369E	Tail Rotor Gearbox Attach Bolts	Aerometals STC SH4801NM

Note 1: This AD applies to each helicopter identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For helicopters that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (c) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and if the unsafe condition has not

been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent loss of the tail rotor gearbox due to attaching bolts of inadequate grip length and subsequent loss of control of the helicopter, accomplish the following:

- (a) Within 25 hours time-in-service (TIS):
 - (1) For each tail rotor gearbox attaching bolt (bolt):
 - (i) Determine the part number (P/N).
 - (ii) If the P/N cannot be determined or if the bolt is not P/N NAS1304–26, before

further flight, replace the bolt with bolt, P/N NAS1304–26.
(iii) Torque the bolt to 100–110 in-lbs and apply a slippage mark.
(2) Remove the tailboom control rod and determine the number of bolt threads protruding from each nutplate on the internal surface of the aft tailboom frame casting, P/N 369D23503, as shown in Figure 1 of this AD. At least one thread must protrude. If more than four threads protrude, add an additional washer, P/N AN960D416, under the bolt head. Torque the bolt to 100–110 in-lbs, and reapply a slippage mark. See Figure 1: