

will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding a new airworthiness directive to read as follows:

2003–04–14 Bell Helicopter Textron

Canada: Amendment 39–13063. Docket No. 2002–SW–19–AD.

Applicability: Model 427 helicopters, serial numbers 56001 through 56031, with hydraulic solenoid tee fitting, part number (P/N) AS1003W060404, installed, certificated in any category.

Note 1: This AD applies to each helicopter identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For helicopters that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (b) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required at the next hydraulic filter and fluid replacement or within 30 days, whichever occurs first, unless accomplished previously.

To prevent restricted flow of hydraulic fluid to the flight control hydraulic actuators resulting in loss of hydraulic control, excessive stiffness in the flight controls, and a subsequent forced landing of the helicopter, accomplish the following:

(a) Replace the hydraulic solenoid tee fitting (tee fitting), P/N AS1003W060404, and

tubes, P/Ns 427–080–058–101 and 427–080–003–101, with union, P/N AS5230W0606, tee fitting, P/N NAS1763W060404, and tubes, P/Ns 427–080–069–101 and 427–080–068–101, in accordance with the Accomplishment Instructions in Bell Helicopter Textron Alert Service Bulletin No. 427–01–02, dated August 20, 2001.

(b) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Regulations Group, Rotorcraft Directorate, FAA. Operators shall submit their requests through an FAA Principal Maintenance Inspector, who may concur or comment and then send it to the Manager, Regulations Group.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Regulations Group.

(c) Special flight permits may be issued in accordance with 14 CFR 21.197 and 21.199 to operate the helicopter to a location where the requirements of this AD can be accomplished.

(d) The replacements shall be done in accordance with the Accomplishment Instructions in Bell Helicopter Textron Alert Service Bulletin No. 427–01–02, dated August 20, 2001. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Bell Helicopter Textron Canada, 12,800 Rue de l'Avenir, Mirabel, Quebec J7J1R4, telephone (450) 437–2862 or (800) 363–8023, fax (450) 433–0272. Copies may be inspected at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(e) This amendment becomes effective on April 4, 2003.

Note 3: The subject of this AD is addressed in Transport Canada (Canada) AD No. CF–2002–11, dated January 31, 2002.

Issued in Fort Worth, Texas, on February 14, 2003.

David A. Downey,

Manager, Rotorcraft Directorate, Aircraft Certification Service.

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2002–NM–15–AD; Amendment 39–13069; AD 2003–04–20]

RIN 2120–AA64

Airworthiness Directives; Raytheon Model DH.125, HS.125, and BH.125 Series Airplanes; Model BAe.125 Series 800A, 800A (C–29A), 800A (U–125), 800B, 1000A, and 1000B Airplanes; and Model Hawker 800, 800 (including variant U–125A), 1000, and 800XP Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment supersedes an existing airworthiness directive (AD), applicable to certain Raytheon Model DH.125, HS.125, BH.125, and BAe.125 (U–125 and C–29A) series airplanes; and Model Hawker 800, Hawker 800 (including variant U–125A), Hawker 800XP, and Hawker 1000 airplanes; that currently requires an inspection for cracking or corrosion of the cylinder head lugs of the main landing gear (MLG) actuator and follow-on/corrective actions. This amendment expands the applicability of the existing AD to add an airplane model and further clarify the applicability and, for certain airplanes, to clarify the compliance time of the inspection requirements. This action is necessary to prevent separation of the cylinder head lugs, which could prevent the MLG from extending and result in a partial gear-up landing. This action is intended to address the identified unsafe condition.

DATES: Effective April 4, 2003.

The incorporation by reference of a certain publication, as listed in the regulations, was approved previously by the Director of the Federal Register as of October 3, 2001 (66 FR 45575, August 29, 2001).

ADDRESSES: The service information referenced in this AD may be obtained from Raytheon Aircraft Company, Department 62, PO Box 85, Wichita, Kansas 67201–0085. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, Wichita Aircraft Certification Office, 1801 Airport Road, Room 100, Wichita, Kansas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.